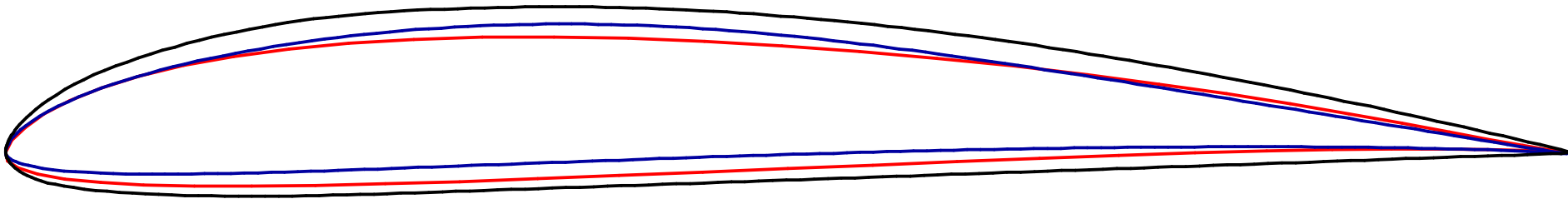


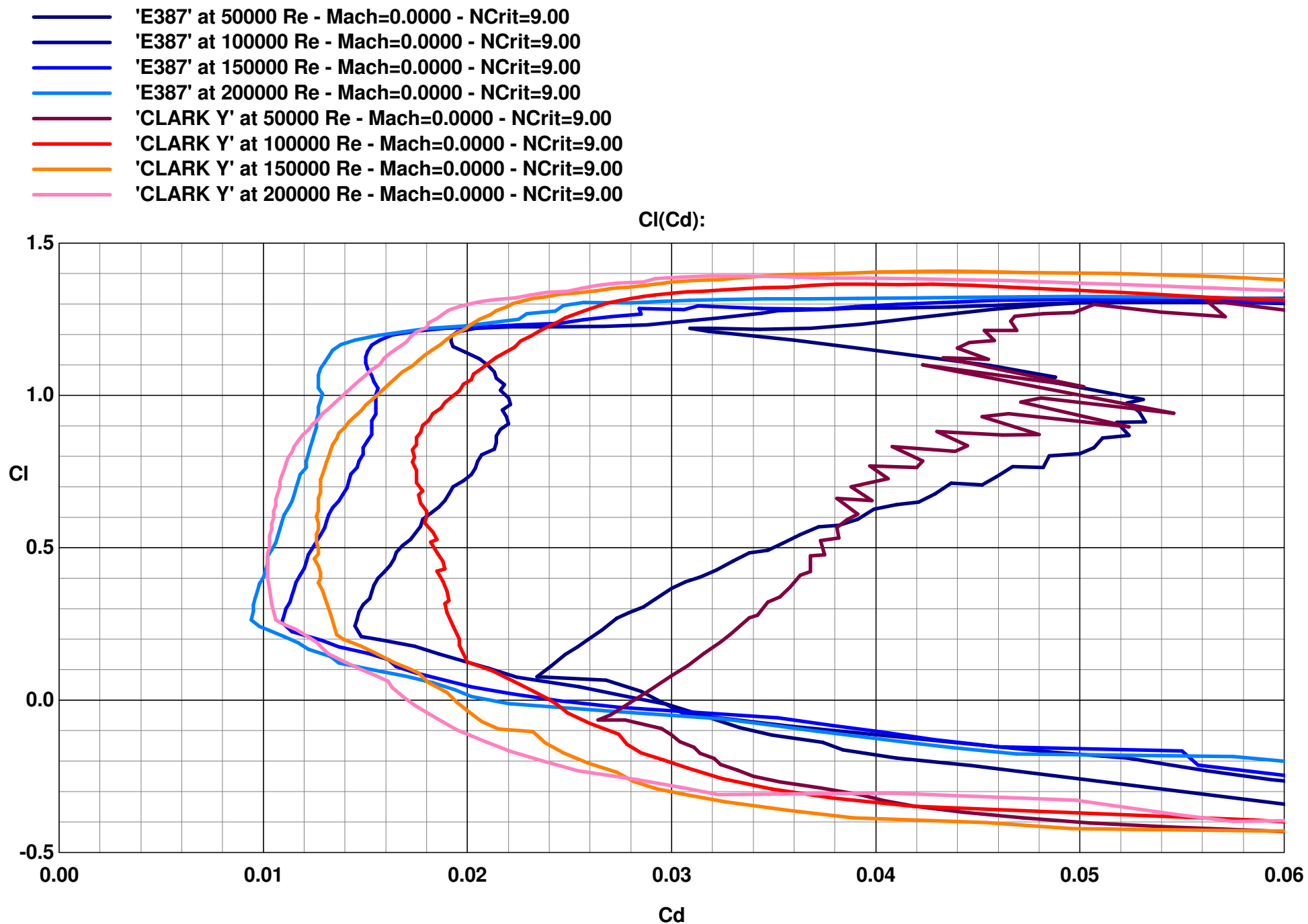
Airfoils in comparison :

SD7037 = 

E387 = 

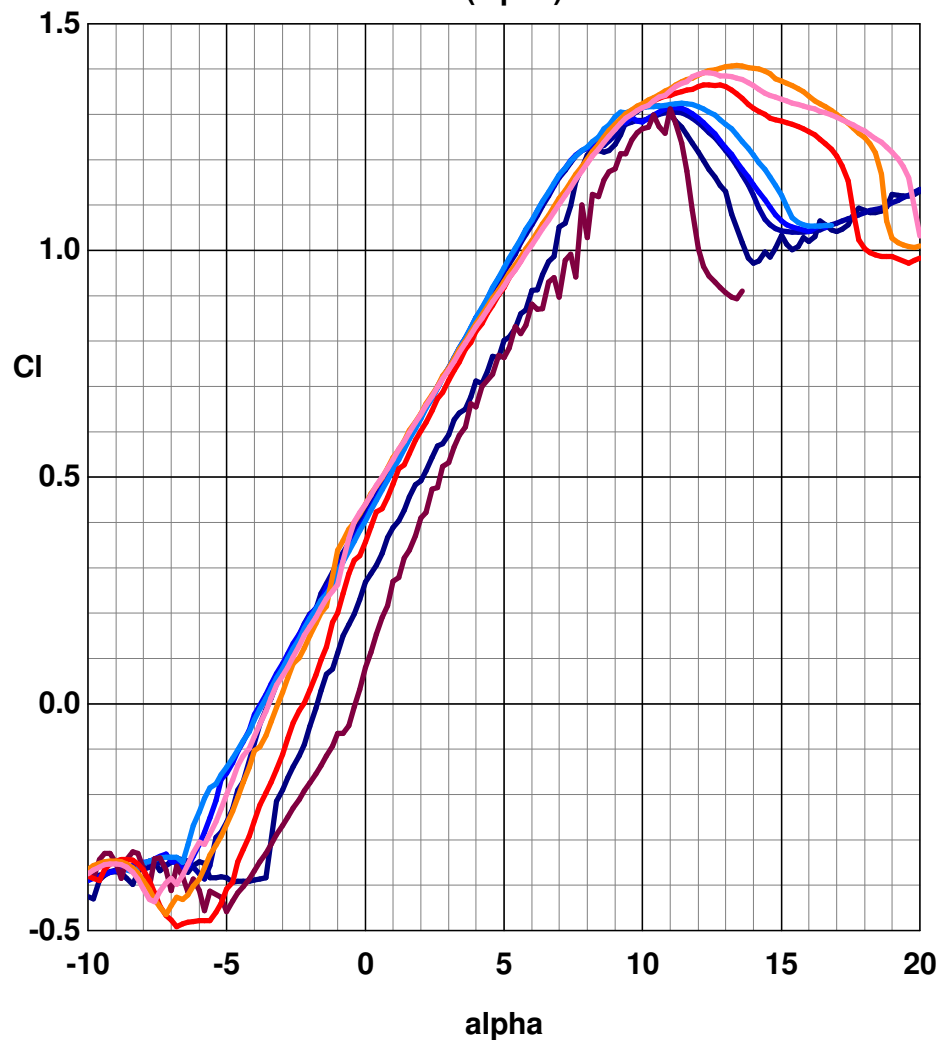
CLARK Y = 



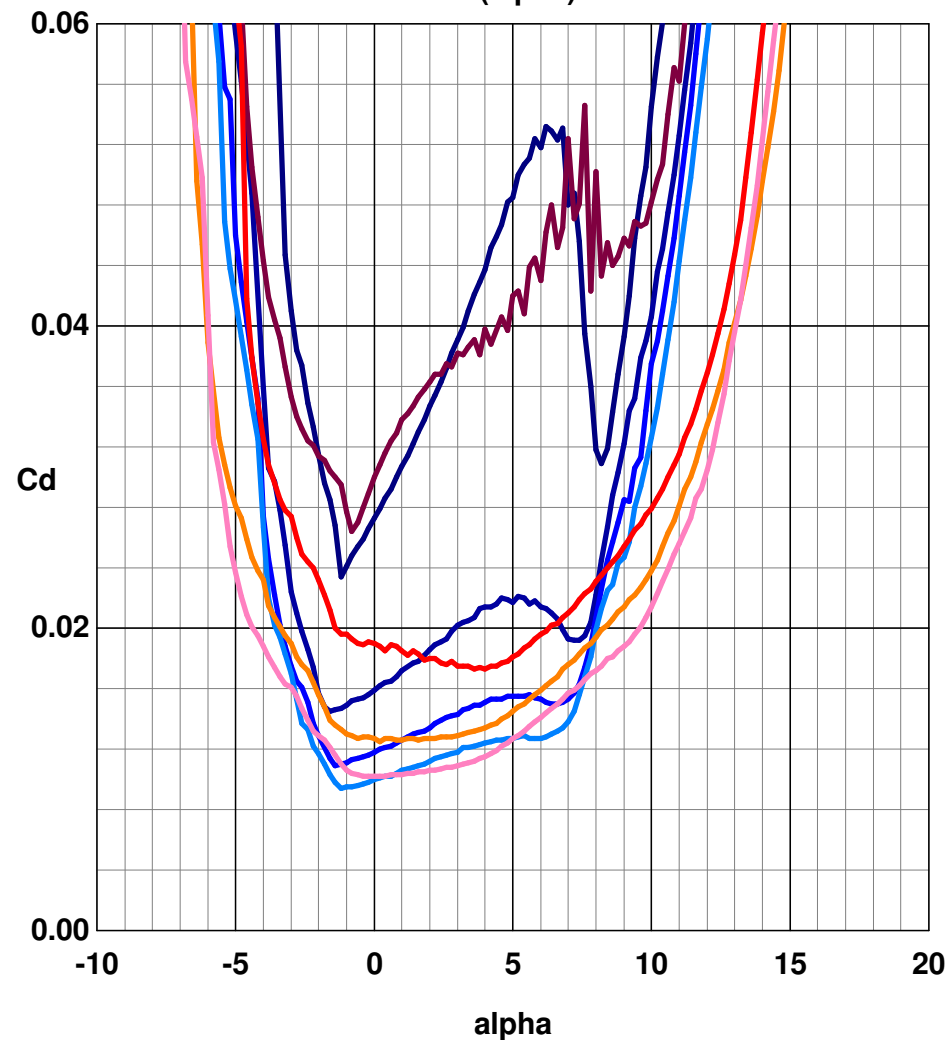


- 'E387' at 50000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 100000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 150000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 200000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 50000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 100000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 150000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 200000 Re - Mach=0.0000 - NCrit=9.00

Cl(alpha):

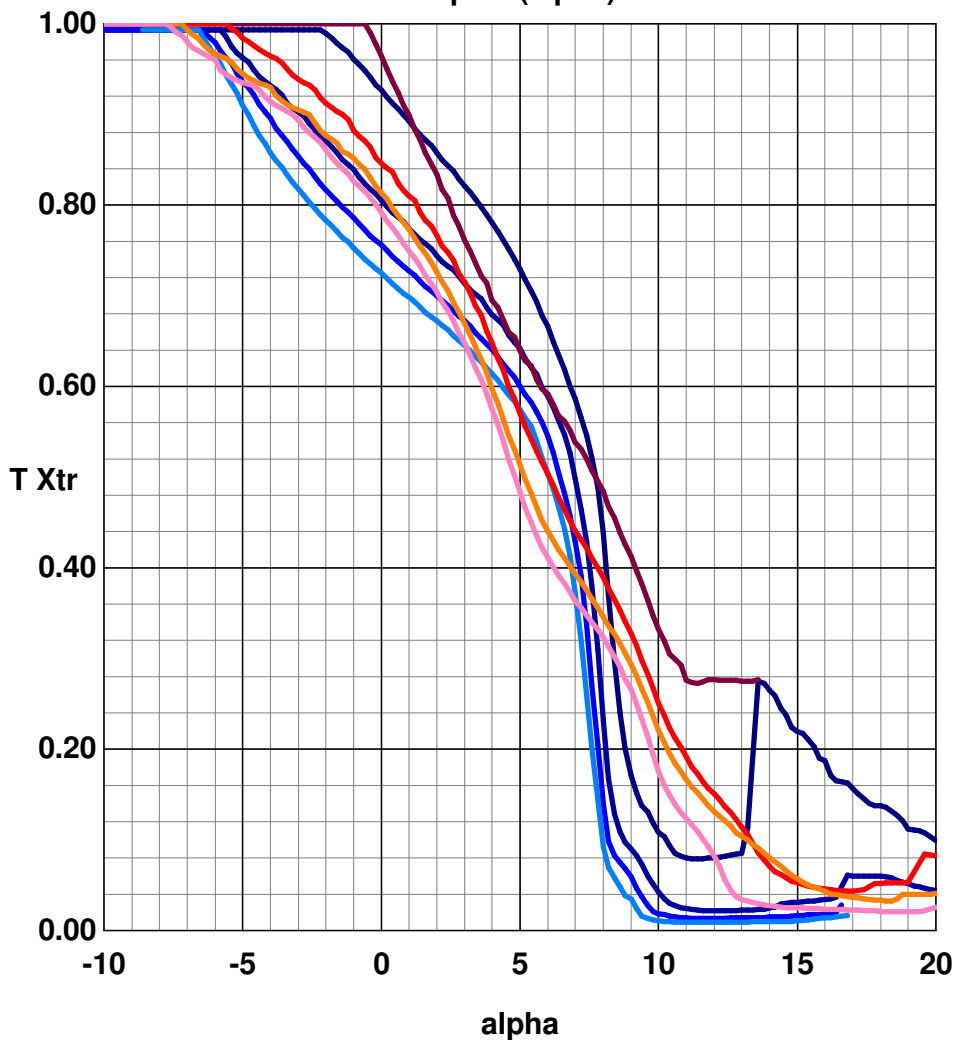


Cd(alpha):



- 'E387' at 50000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 100000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 150000 Re - Mach=0.0000 - NCrit=9.00
- 'E387' at 200000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 50000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 100000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 150000 Re - Mach=0.0000 - NCrit=9.00
- 'CLARK Y' at 200000 Re - Mach=0.0000 - NCrit=9.00

Top Xtr(alpha):



Bottom Xtr(alpha):

